

**REPORT OF AIRCRAFT WEIGHING**

Report N.

**220L/19**

Reg. Marks	Type	Serial Number	C. N. n°	Operator	Place and Date
N985DS	V1.0	1027	-	-	NAPLES 10/12/19
Division		N. Passengers		Without de-icing the wings and tail	
-		0			

 Reason for Weighing: DETERMING WEIGHT AND CENTER OF GRAVITY AFTER CONSTRUCTION

 Measuring instrument used: CAPTELS ORA10 MVN

 Position of airplane: LEVELLED

 Weighing point used: NOSE WHEEL AND MAIN WHEELS

 Reference plane of longitudinal distance: VERTICAL TANGENT LEADING EDGE WING

 Mean aerodynamic chord (MAC) Length **L 53,5 in**

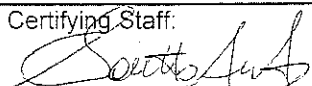
 Datum **Y= 0 in**

Weighing point	Net Weight Lb	Arm in	Moment Lb x in
Left	709,8876	26,38	18726,83
Right	692,2507	26,38	18261,57
Nose	313,497	-42,52	-13329,89
<b>TOTAL</b>	<b>A 1715,6353</b>		<b>E 23658,52</b>
Weights to be added	<b>B 15,87 =</b>	<b>25,59</b>	<b>F 406,11 =</b>
<b>A + B ⇒</b>	<b>1731,5053</b>	<b>E + F ⇒</b>	<b>24064,63 -</b>
Weights to be subtracted	<b>C 0,00 =</b>	<b>0,65</b>	<b>H 0,00 =</b>
<b>BASIC EMPTY WEIGHT G=A+B-C</b>	<b>1731,5</b>	<b>BASIC EMPTY WEIGHT MOMENT M=E+F-H</b>	<b>24064,629</b>

 Comments : The basic empty weight is related to the aircraft configuration Standard.  
 Weight to point A includes engine oil and hydraulic fluids of all system

Basic Empty weight Arm <b>X = M/G =</b>	<b>13,898</b>	% MAC = <b>(X-Y)/L =</b>	<b>26,0%</b>
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Certifying Staff:



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